

Design, Analysis and Research Corporation (DARcorporation)

ERRATA: Airplane Design Part I
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- line 6 from bottom* replace 1,000 nm with 1,000 sm
- page 61, Table 2.19* Take-off and Landing: groundrun of less than 2,400 ft.
- page 69, Eqn. (2.23)* $D = (W_{PL} + W_{CREW}) + W_{PL_{exp}}$
Where $W_{p_{exp}}$ is the weight of the expended payload.
(i.e., Missiles, bombs, etc.)
- page 98, line 8 from bottom* replace $C_{LTO_{max}}$ by $C_{L_{max}_{TO}}$
- page 106, line 2* replace four factors: with five factors:
- page 115, Eqn. (3.18)* $V_A = 1.1 V_{s_{PA}}$
- page 150, Eqn. (3.32)* $RC_h = RC_0(1 - h / h_{abs})$
- page 152, Eqn. (3.38)*
$$\sin\gamma = \frac{T}{W} \left[P_{dl} - \sqrt{P_{dl}^2 - P_{dl} + \left(1 + \left(\frac{L}{D}\right)^2\right)^{-1} \left(\frac{T}{W}\right)^{-2}} \right]$$
- page 186, Section 3.7.4.2*
line 3 ...groundrun as < 2,400 ft.
- line 7* $S_L = 1.9 \times 2,400 =$
 $= 4,560 \text{ ft.}$
- From Figure 3.16* $S_L = 4,500/0.6 = 7,600 \text{ ft.}$
- From Figure 3.17 this yields:* $V_A^2 = 25,000 \text{ kts}^2$
 $V_A = \{21,200(1.3/1.2)^2\}^{1/2} = 158 \text{ kts}$ should be
 $V_A = \{25,000\}^{1/2} = 158 \text{ kts}$

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